EE 565: Position, Navigation and Timing Error Mechanization (NAV)

Aly El-Osery Kevin Wedeward

Electrical Engineering Department, New Mexico Tech Socorro, New Mexico, USA

In Collaboration with
Stephen Bruder
Electrical and Computer Engineering Department
Embry-Riddle Aeronautical Univesity
Prescott, Arizona, USA

April 9, 2023

NAV Attitude Error



$$\dot{C}_b^n = C_b^n(\Omega_{ib}^b - \Omega_{in}^b) = \frac{d}{dt} \left[(\mathcal{I} + [\delta \vec{\psi}_{nb}^n \times]) \hat{C}_b^n \right] =$$



$$\dot{C}_b^n = C_b^n(\Omega_{ib}^b - \Omega_{in}^b) = \frac{d}{dt} \left[(\mathcal{I} + [\delta \vec{\psi}_{nb}^n \times]) \hat{C}_b^n \right] =$$

$$(\mathcal{I} + [\delta\vec{\psi}_{nb}^{n} \times])\hat{C}_{b}^{n}\Omega_{nb}^{b} = [\delta\vec{\psi}_{nb}^{n} \times]\hat{C}_{b}^{n} + (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n} \times])\hat{C}_{b}^{n} =$$



$$\dot{C}_b^n = C_b^n (\Omega_{ib}^b - \Omega_{in}^b) = \frac{d}{dt} \left[(\mathcal{I} + [\delta \vec{\psi}_{nb}^n \times]) \hat{C}_b^n \right] =$$

$$(\mathcal{I} + [\delta \vec{\psi}_{nb}^{n} \times]) \hat{C}_{b}^{n} \Omega_{nb}^{b} = [\delta \vec{\psi}_{nb}^{n} \times] \hat{C}_{b}^{n} + (\mathcal{I} + [\delta \vec{\psi}_{nb}^{n} \times]) \hat{C}_{b}^{n} =$$

$$\approx (\mathcal{I} + [\delta \vec{\psi}_{nb}^{n} \times]) \hat{C}_{b}^{n} (\hat{\Omega}_{nb}^{b} + \delta \Omega_{ib}^{b} - \delta \Omega_{in}^{b})$$



$$\dot{C}_b^n = C_b^n(\Omega_{ib}^b - \Omega_{in}^b) = \frac{d}{dt} \left[(\mathcal{I} + [\delta \vec{\psi}_{nb}^n \times]) \hat{C}_b^n \right] =$$

$$\begin{aligned} & (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}\Omega_{nb}^{b} = [\delta\dot{\vec{\psi}}_{nb}^{n}\times]\hat{C}_{b}^{n} + (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\dot{\hat{C}}_{b}^{n} = \\ & \approx (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}(\hat{\Omega}_{nb}^{b} + \delta\Omega_{ib}^{b} - \delta\Omega_{in}^{b}) \\ & \approx (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}\hat{\Omega}_{nb}^{b} + \hat{C}_{b}^{n}(\delta\Omega_{ib}^{b} - \delta\Omega_{in}^{b}) \\ & \approx (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}\hat{\Omega}_{nb}^{b} + \hat{C}_{b}^{n}(\delta\Omega_{ib}^{b} - \delta\Omega_{in}^{b}) \\ & \qquad \qquad \cdot \cdot [\delta\vec{\psi}_{nb}^{n}\times]\delta\Omega_{nb}^{b} \approx 0 \end{aligned}$$

NAV Attitude Error



$$\dot{C}_b^n = C_b^n(\Omega_{ib}^b - \Omega_{in}^b) = \frac{d}{dt} \left[(\mathcal{I} + [\delta \vec{\psi}_{nb}^n \times]) \hat{C}_b^n \right] =$$

$$(\mathcal{I} + [\delta \vec{\psi}_{nb}^{n} \times]) \hat{C}_{b}^{n} \Omega_{nb}^{b} = [\delta \dot{\vec{\psi}}_{nb}^{n} \times] \hat{C}_{b}^{n} + (\mathcal{I} + [\delta \vec{\psi}_{nb}^{n} \times]) \hat{C}_{b}^{n}$$

$$\approx (\mathcal{I} + [\delta \vec{\psi}_{nb}^{n} \times]) \hat{C}_{b}^{n} (\hat{\Omega}_{nb}^{b} + \delta \Omega_{ib}^{b} - \delta \Omega_{in}^{b})$$

$$\approx (\mathcal{I} + [\delta \vec{\psi}_{nb}^{n} \times]) \hat{C}_{b}^{n} \hat{\Omega}_{nb}^{b} + \hat{C}_{b}^{n} (\delta \Omega_{ib}^{b} - \delta \Omega_{in}^{b})$$

$$\approx (\mathcal{I} + [\delta \vec{\psi}_{nb}^{n} \times]) \hat{C}_{b}^{n} \hat{\Omega}_{nb}^{b} + \hat{C}_{b}^{n} (\delta \Omega_{ib}^{b} - \delta \Omega_{in}^{b})$$



$$\dot{C}_b^n = C_b^n (\Omega_{ib}^b - \Omega_{in}^b) = \frac{d}{dt} \left[(\mathcal{I} + [\delta \vec{\psi}_{nb}^n \times]) \hat{C}_b^n \right] =$$

$$\begin{split} &(\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}\Omega_{nb}^{b} = [\delta\dot{\psi}_{nb}^{n}\times]\hat{C}_{b}^{n} + (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\dot{\hat{C}}_{b}^{n} = \\ &\approx (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}(\hat{\Omega}_{nb}^{b} + \delta\Omega_{ib}^{b} - \delta\Omega_{in}^{b}) \\ &\approx (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}\hat{\Omega}_{nb}^{b} + \hat{C}_{b}^{n}(\delta\Omega_{ib}^{b} - \delta\Omega_{in}^{b}) \end{split}$$

$$[\delta \dot{\vec{\psi}}_{nb}^{n} \times] = \hat{C}_{b}^{n} (\delta \Omega_{nb}^{b} - \delta \Omega_{in}^{b}) \hat{C}_{n}^{b} = [\hat{C}_{b}^{n} (\delta \vec{\omega}_{ib}^{b} - \delta \vec{\omega}_{in}^{b}) \times]$$
(1)

Attitude •0000 NAV Velocity Error

Nav Position Err

Summar



$$\dot{C}_b^n = C_b^n(\Omega_{ib}^b - \Omega_{in}^b) = \frac{d}{dt} \left[(\mathcal{I} + [\delta \vec{\psi}_{nb}^n \times]) \hat{C}_b^n \right] =$$

$$\begin{split} &(\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}\Omega_{nb}^{b} = [\delta\dot{\psi}_{nb}^{n}\times]\hat{C}_{b}^{n} + (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\dot{\hat{C}}_{b}^{n} = \\ &\approx (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}(\hat{\Omega}_{nb}^{b} + \delta\Omega_{ib}^{b} - \delta\Omega_{in}^{b}) \\ &\approx (\mathcal{I} + [\delta\vec{\psi}_{nb}^{n}\times])\hat{C}_{b}^{n}\hat{\Omega}_{nb}^{b} + \hat{C}_{b}^{n}(\delta\Omega_{ib}^{b} - \delta\Omega_{in}^{b}) \end{split}$$

$$[\delta \vec{\psi}_{nb}^{n} \times] = \hat{C}_{b}^{n} (\delta \Omega_{nb}^{b} - \delta \Omega_{in}^{b}) \hat{C}_{n}^{b} = [\hat{C}_{b}^{n} (\delta \vec{\omega}_{ib}^{b} - \delta \vec{\omega}_{in}^{b}) \times]$$

$$(1)$$

$$\delta \dot{\vec{\psi}}_{nb}^{n} = \hat{C}_{b}^{n} (\delta \omega_{ib}^{b} - \delta \vec{\omega}_{in}^{b}) \tag{2}$$

NAV Attitude Error (cont.)



Recall that

Computing $\delta \vec{\omega}_{in}^{b}$

$$\vec{\omega}_{in}^{b} = C_n^b \vec{\omega}_{in}^n$$

Expressing the above equation in terms of estimates we get

$$\hat{\vec{\omega}}_{in}^{b} + \delta \vec{\omega}_{in}^{b} = \hat{C}_{n}^{b} (\mathcal{I} - [\delta \vec{\psi}_{nb}^{n} \times]) (\hat{\vec{\omega}}_{in}^{n} + \delta \vec{\omega}_{in}^{n})$$
$$\delta \vec{\omega}_{in}^{b} \approx \hat{C}_{n}^{b} (\delta \vec{\omega}_{in}^{n} - [\delta \vec{\psi}_{nb}^{n} \times] \hat{\vec{\omega}}_{in}^{n})$$

Substituting this result in Equation 2

$$\delta \dot{\vec{\psi}}_{nb}^{n} = -\hat{\Omega}_{in}^{n} \delta \vec{\psi}_{nb}^{n} + \hat{C}_{b}^{n} \delta \vec{\omega}_{ib}^{b} - \delta \vec{\omega}_{in}^{n}$$
(3)

3 / 23



Using Taylor series and retaining the first order terms

$$\delta \vec{\omega}_{in}^{n} = \frac{\partial \hat{\vec{\omega}}_{in}^{n}}{\partial \hat{\vec{r}}_{eb}^{n}} \delta \vec{r}_{eb}^{n} + \frac{\partial \hat{\vec{\omega}}_{in}^{n}}{\partial \hat{\vec{v}}_{eb}^{n}} \delta \vec{v}_{eb}^{n}$$

where $\vec{r}_{eb}^n = [L_b, \lambda_b, h_b]^T$ and

$$\hat{\vec{\omega}}_{in}^{n} = \hat{\vec{\omega}}_{ie}^{n} + \hat{\vec{\omega}}_{en}^{n} = \begin{pmatrix} \omega_{ie} \cos \hat{L}_b + \frac{\hat{\vec{v}}_{eb,E}^{n}}{R_E(\hat{L}_b) + \hat{h}_b} \\ -\frac{\hat{\vec{v}}_{eb,N}^{n}}{R_N(\hat{L}_b) + \hat{h}_b} \\ -\omega_{ie} \sin \hat{L}_b - \frac{\hat{\vec{v}}_{eb,E}^{n} \tan \hat{L}_b}{R_E(\hat{L}_b) + \hat{h}_b} \end{pmatrix}$$

NAV Attitude Error (cont.)

Define terms for simplicity



$$-\,\hat{\vec{\Omega}}_{in}^{\,n} = -[(\hat{\vec{\omega}}_{ie}^{\,n} + \hat{\vec{\omega}}_{en}^{\,n}) \times] = F_{\psi\psi}$$

$$-\frac{\partial \hat{\vec{\omega}} \frac{n}{in}}{\partial \hat{\vec{V}} \frac{n}{eb}} = \begin{pmatrix} 0 & -\frac{1}{R_E(\hat{\ell}_b) + \hat{h}_b} & 0\\ \frac{1}{R_N(\hat{\ell}_b) + \hat{h}_b} & 0 & 0\\ 0 & \frac{\tan \hat{\ell}_b}{2\pi} & 0 \end{pmatrix} = F_{\psi v}$$

$$-\frac{\partial \hat{\omega}_{in}^{n}}{\partial \hat{r}_{eb}^{n}} = \begin{pmatrix} \omega_{ie} \sin \hat{L}_{b} & 0 & \frac{\hat{v}_{eb,E}^{n}}{(R_{E}(\hat{L}_{b}) + \hat{h}_{b})^{2}} \\ 0 & 0 & -\frac{\hat{v}_{eb,B}^{n}}{(R_{N}(\hat{L}_{b}) + \hat{h}_{b})^{2}} \\ \omega_{ie} \cos \hat{L}_{b} + \frac{\hat{v}_{eb,E}^{n}}{(R_{E}(\hat{L}_{b}) + \hat{h}_{b}) \cos^{2} \hat{L}_{b}} & 0 & -\frac{\hat{v}_{eb,E}^{n} \tan \hat{L}_{b}}{(R_{E}(\hat{L}_{b}) + \hat{h}_{b})^{2}} \end{pmatrix} = F_{\psi r}$$

(5)

(4)

NAV Attitude Error (cont.)

Final Expression



$$\delta \dot{\vec{\psi}}_{nb}^{n} = F_{\psi\psi} \delta \vec{\psi}_{nb}^{n} + F_{\psi\nu} \delta \vec{v}_{eb}^{n} + F_{\psi r} \delta \vec{r}_{eb}^{n} + \hat{C}_{b}^{n} \delta \vec{\omega}_{ib}^{b}$$

$$(7)$$

where $F_{\psi\psi}$, $F_{\psi v}$ and $F_{\psi r}$ are defined by Equations 4, 5 and 6, respectively.



$$\dot{\vec{v}}_{eb}^{n} = C_{b}^{n} \vec{f}_{ib}^{b} + \vec{g}_{b}^{n} - (\Omega_{en}^{n} + 2\Omega_{ie}^{n}) \vec{v}_{eb}^{e}$$
(8)

$$\dot{\hat{v}}_{eb}^{n} = \hat{C}_{b}^{n} \hat{f}_{ib}^{b} + \hat{g}_{b}^{n} - (\hat{\Omega}_{en}^{n} + 2\hat{\Omega}_{ie}^{n}) \hat{v}_{eb}^{n}
= (\mathcal{I} - [\delta \vec{\psi}_{nb}^{n} \times]) C_{b}^{n} (\vec{f}_{ib}^{b} - \delta \vec{f}_{ib}^{b}) + \hat{g}_{b}^{n} - (\hat{\Omega}_{en}^{n} + 2\hat{\Omega}_{ie}^{n}) \hat{v}_{eb}^{n}$$
(9)

Computing the $\delta \vec{v}_{eb}^{n}$ we obtain

$$\begin{split} \delta \dot{\vec{v}}_{eb}^{n} &= \dot{\vec{v}}_{eb}^{n} - \dot{\hat{\vec{v}}}_{eb}^{n} \\ &= [\delta \vec{\psi}_{nb}^{n} \times] C_{b}^{n} \vec{f}_{ib}^{b} + \hat{C}_{b}^{n} \delta \vec{f}_{ib}^{b} + \delta \vec{g}_{b}^{n} + \\ &- (\hat{\Omega}_{en}^{n} + 2\hat{\Omega}_{ie}^{n}) \delta \vec{v}_{eb}^{n} - (\delta \Omega_{en}^{n} + 2\delta \Omega_{ie}^{n}) \hat{\vec{v}}_{eb}^{n} \end{split}$$

Velocity Cont.

$-(\hat{\Omega}_{en}^n+2\hat{\Omega}_{ie}^n)$



The term $-(\hat{\Omega}_{en}^n + 2\hat{\Omega}_{ie}^n)$ is dervied as

$$-\left[(\hat{\vec{\omega}}_{en}^{n}+2\hat{\vec{\omega}}_{ie}^{n})\times\right] = \begin{pmatrix} 0 & -2\omega_{ie}\sin\hat{L}_{b} - \frac{\hat{\vec{v}}_{eb,E}^{n}\tan\hat{L}_{b}}{R_{E}(\hat{L}_{b})+\hat{h}_{b}} & \frac{\hat{\vec{v}}_{eb,E}^{n}}{R_{N}(\hat{L}_{b})+\hat{h}_{b}} \\ 2\omega_{ie}\sin\hat{L}_{b} + \frac{\hat{\vec{v}}_{eb,E}^{n}\tan\hat{L}_{b}}{R_{E}(\hat{L}_{b})+\hat{h}_{b}} & 0 & 2\omega_{ie}\cos\hat{L}_{b} + \frac{\hat{\vec{v}}_{eb,E}^{n}}{R_{E}(\hat{L}_{b})+\hat{h}_{b}} \\ -\frac{\hat{\vec{v}}_{eb,B}^{n}}{R_{N}(\hat{L}_{b})+\hat{h}_{b}} & -2\omega_{ie}\cos\hat{L}_{b} - \frac{\hat{\vec{v}}_{eb,E}^{n}}{R_{E}(\hat{L}_{b})+\hat{h}_{b}} & 0 \end{pmatrix} = F_{\Omega\Omega}$$

$$(10)$$

Velocity Cont.

$$-(\delta\hat{\Omega}_{en}^n+2\delta\hat{\Omega}_{ie}^n)\hat{\vec{v}}_{eb}^n$$



(11)

The term $-(\delta \hat{\Omega}_{en}^n + 2\delta \hat{\Omega}_{ie}^n)\hat{\vec{v}}_{eb}^n$ is dervied as

$$-(\delta\hat{\Omega}_{en}^{n}+2\delta\hat{\Omega}_{ie}^{n})\hat{\vec{v}}_{eb}^{n}=\frac{\partial F_{\Omega\Omega}}{\partial\hat{\vec{r}}_{eb}^{n}}\delta\vec{r}_{eb}^{n}\hat{\vec{v}}_{eb}^{n}+\frac{\partial F_{\Omega\Omega}}{\partial\hat{\vec{v}}_{eb}^{n}}\delta\vec{v}_{eb}^{n}\hat{\vec{v}}_{eb}^{n}$$

Nav Position Error Summar

Velocity Cont. Gravity Error



$$\delta \vec{g}_b^n pprox \frac{2g_0(\hat{L}_b)}{r_{eS}^e(\hat{L}_b)} \delta h_b$$

12)

Attitude

NAV Velocity Error

Nav Position E

O

Velocity Error Cont.

After tons of algebra



$$\delta \dot{\vec{v}}_{eb}^{n} = F_{v\psi} \delta \vec{\psi}_{nb}^{n} + F_{vv} \delta \vec{v}_{eb}^{n} + F_{vr} \delta \vec{r}_{eb}^{n} + \hat{C}_{b}^{n} \delta \vec{f}_{ib}^{b}$$

$$(13)$$

where $F_{v\psi}$, F_{vv} and F_{vr} are defined by Equations 14, 15 and 16, respectively.

Velocity Cont.

$F_{v\psi}$, F_{vv} and F_{vr}



$$F_{\nu\psi} = -\left[(\hat{C}_b^{\eta} \hat{f}_{b}^{\theta}) \times \right] \tag{14}$$

$$\mathsf{F}_{vv} = \begin{pmatrix} \frac{v_{ob,D}^n}{(R_N(\tilde{L}_b) + \tilde{h}_b)} & -\frac{2\tilde{v}_{ob,E}^n \tan \hat{L}_b}{(R_E(\tilde{L}_b) + \tilde{h}_b)} - 2\omega_{ie} \sin \hat{L}_b & \frac{\tilde{v}_{ob,N}^n}{R_N(\tilde{L}_b) + \tilde{h}_b} \\ \\ \frac{\tilde{v}_{ob,E}^n \tan \hat{L}_b}{(R_E(\tilde{L}_b) + \tilde{h}_b)} + 2\omega_{ie} \sin \hat{L}_b & \frac{\tilde{v}_{ob,N}^n \tan \hat{L}_b + \tilde{v}_{ob,D}^n}{R_N(\tilde{L}_b) + \tilde{h}_b} & \frac{\tilde{v}_{ob,E}^n}{(R_E(\tilde{L}_b) + \tilde{h}_b)} + 2\omega_{ie} \cos \hat{L}_b \\ \\ -\frac{2\tilde{v}_{ob,N}^n}{R_N(\tilde{L}_b) + \tilde{h}_b} & -\frac{2\tilde{v}_{ob,E}^n}{(R_E(\tilde{L}_b) + \tilde{h}_b)} - 2\omega_{ie} \cos \hat{L}_b & 0 \end{pmatrix}$$

$$\mathsf{F}_{vr} = \begin{pmatrix} -\frac{(\hat{v}_{eb,E}^n)^2 \sec^2 \hat{L}_b}{R_E(\hat{L}_b) + \hat{h}_b} - 2\hat{v}_{eb,E}^n \omega_{ie} \cos \hat{L}_b & 0 & \frac{(\hat{v}_{eb,E}^n)^2 \tan \hat{L}_b}{(R_E(\hat{L}_b) + \hat{h}_b)^2} - \frac{\hat{v}_{eb,N}^n \hat{v}_{eb,D}^n}{(R_N(\hat{L}_b) + \hat{h}_b)^2} \\ \frac{\hat{v}_{eb,N}^n \hat{v}_{eb,E}^n \sec^2 \hat{L}_b}{R_E(\hat{L}_b) + \hat{h}_b} + 2\hat{v}_{eb,N}^n \omega_{ie} \cos \hat{L}_b - 2\hat{v}_{eb,D}^n \omega_{ie} \sin \hat{L}_b & 0 & -\frac{\hat{v}_{eb,N}^n \hat{v}_{eb,E}^n \tan \hat{L}_b + \hat{v}_{eb,E}^n \hat{v}_{eb,D}^n}{(R_E(\hat{L}_b) + \hat{h}_b)^2} \\ 2\hat{v}_{eb,E}^n \omega_{ie} \sin \hat{L}_b & 0 & \left(\frac{(\hat{v}_{eb,E}^n)^2}{(R_E(\hat{L}_b) + \hat{h}_b)^2} + \frac{(\hat{v}_{eb,N}^n)^2}{(R_N(\hat{L}_b) + \hat{h}_b)^2} - \frac{2g_0(\hat{L}_b)}{r_{ea}^n(\hat{L}_b)}\right) \end{pmatrix}$$

$$(16)$$

(15)



$$\dot{\vec{r}}_{eb}^{n} = \begin{pmatrix} \frac{\hat{\vec{v}}_{eb,N}^{n}}{R_{N}(\hat{L}_{b}) + \hat{h}_{b}} \\ \frac{\hat{\vec{v}}_{eb,E}^{n}}{\cos \hat{L}_{b}(R_{E}(\hat{L}_{b}) + \hat{h}_{b})} \\ -\hat{\vec{v}}_{eb,D}^{n} \end{pmatrix}$$
(17)

Computing $\delta \vec{r}_{eb}^n$ using taylor series expansion and retaining only the first order terms

$$\delta \vec{r}_{eb}^{n} = F_{r\psi} \delta \vec{\psi}_{nb}^{n} + F_{rv} \delta \vec{v}_{eb}^{n} + F_{rr} \delta \vec{r}_{eb}^{n}$$
(18)

where F_{rv} , F_{rv} and F_{rr} are defined by Equations 19, 20 and 21, respectively.

F_{rab} , F_{rv} and F_{rr}



$$F_{r\psi} = 0_{3\times3}$$

$$\mathsf{F}_{r \psi} = \mathsf{0}_{3 imes 3}$$

$$\mathsf{F}_{r v} = egin{pmatrix} rac{1}{R_N(\hat{L}_b) + \hat{h}_b} & 0 & 0 \\ 0 & rac{1}{(R_E(\hat{L}_b) + \hat{h}_b)\cos\hat{L}_b} & 0 \\ 0 & 0 & -1 \end{pmatrix}$$

$$\mathsf{F}_{rr} = \begin{pmatrix} 0 & 0 & -\frac{\hat{v}_{eb,N}^n}{(R_N(\hat{L}_b) + \hat{h}_b)^2} \\ \\ \frac{\hat{v}_{eb,E}^n \tan \hat{L}_b}{(R_N(\hat{L}_b) + \hat{h}_b) \cos \hat{L}_b} & 0 & -\frac{\hat{v}_{eb,E}^n}{(R_N(\hat{L}_b) + \hat{h}_b)^2 \cos \hat{L}_b} \\ \\ 0 & 0 & 0 \end{pmatrix}$$

|Summary - in terms of $\delta ec{f}_{ib}^{\ b}, \delta ec{\omega}_{ib}^{\ b}$



$$\begin{pmatrix}
\delta \dot{\vec{\psi}}_{nb}^{n} \\
\delta \dot{\vec{v}}_{nb}^{n} \\
\delta \dot{\vec{r}}_{nb}^{n}
\end{pmatrix} = \begin{pmatrix}
F_{\psi\psi} & F_{\psi\nu} & F_{\psi r} \\
F_{\nu\psi} & F_{\nu\nu} & F_{\nu r} \\
0_{3\times3} & F_{r\nu} & F_{rr}
\end{pmatrix} \begin{pmatrix}
\delta \vec{\psi}_{nb}^{n} \\
\delta \vec{v}_{nb}^{n} \\
\delta \vec{r}_{nb}^{n}
\end{pmatrix} + \begin{pmatrix}
0 & \hat{C}_{b}^{n} \\
\hat{C}_{b}^{n} & 0 \\
0 & 0
\end{pmatrix} \begin{pmatrix}
\delta \vec{f}_{ib}^{b} \\
\delta \vec{\omega}_{ib}^{b}
\end{pmatrix}$$
(22)

Attitude NAV Velocity Error Nav Position Error Summary



Truth value

$$\vec{x}$$

Measured value

Estimated or computed value

$$\hat{\vec{x}}$$

Error

$$\delta \vec{x} = \vec{x} - \hat{\vec{x}}$$



Nothing above

- Truth value
- Measured value

- Estimated or computed value
- Error







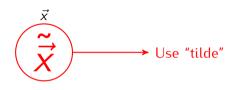
$$\delta \vec{x} = \vec{x} - \hat{\vec{x}}$$



- Truth value
- Measured value

• Estimated or computed value

Error



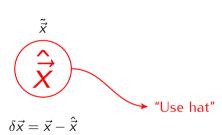


$$\delta \vec{x} = \vec{x} - \hat{\vec{x}}$$



- Truth value
- Measured value
- Estimated or computed value

Error



 \vec{x}



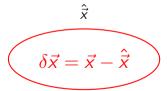
Truth value



Measured value



- Estimated or computed value
- Error





Given a non-linear system $\dot{\vec{x}} = f(\vec{x}, t)$



Given a non-linear system $\dot{\vec{x}} = f(\vec{x}, t)$

Let's assume we have an estimate of \vec{x} , i.e., $\hat{\vec{x}}$ such that $\vec{x} = \hat{\vec{x}} + \delta \vec{x}$

$$\dot{ec{x}}=\dot{ec{x}}+\delta\dot{ec{x}}=f(\hat{ec{x}}+\deltaec{x},t)$$



Given a non-linear system $\vec{x} = f(\vec{x}, t)$

Let's assume we have an estimate of \vec{x} , i.e., $\hat{\vec{x}}$ such that $\vec{x} = \hat{\vec{x}} + \delta \vec{x}$

$$\dot{\vec{x}} = \dot{\vec{x}} + \delta \dot{\vec{x}} = f(\hat{\vec{x}} + \delta \vec{x}, t) \tag{23}$$

Using Taylor series expansion

$$f(\hat{\vec{x}} + \delta \vec{x}, t) = \dot{\hat{\vec{x}}} + \delta \dot{\vec{x}} = f(\hat{\vec{x}}, t) + \frac{\partial f(\vec{x}, t)}{\partial \vec{x}} \Big|_{\vec{x} = \hat{\vec{x}}} \delta \vec{x} + H.O.T$$

$$\approx \dot{\hat{\vec{x}}} + \frac{\partial f(\vec{x}, t)}{\partial \vec{x}} \Big|_{\vec{x} = \hat{\vec{x}}} \delta \vec{x}$$



Given a non-linear system $\vec{x} = f(\vec{x}, t)$

Let's assume we have an estimate of \vec{x} , i.e., $\hat{\vec{x}}$ such that $\vec{x} = \hat{\vec{x}} + \delta \vec{x}$

$$\dot{\vec{x}} = \dot{\vec{x}} + \delta \dot{\vec{x}} = f(\hat{\vec{x}} + \delta \vec{x}, t)$$
 (23)

Using Taylor series expansion

$$f(\hat{\vec{x}} + \delta \vec{x}, t) = \dot{\hat{\vec{x}}} + \delta \dot{\vec{x}} = \underbrace{f(\hat{\vec{x}}, t)}_{\vec{x}} + \underbrace{\frac{\partial f(\vec{x}, t)}{\partial \vec{x}}}_{\vec{x} = \hat{\vec{x}}} \delta \vec{x} + H.O.T$$

$$\approx \dot{\hat{\vec{x}}} + \frac{\partial f(\vec{x}, t)}{\partial \vec{x}} \Big|_{\vec{x} = \hat{\vec{x}}} \delta \vec{x}$$



Given a non-linear system $\vec{x} = f(\vec{x}, t)$

Let's assume we have an estimate of \vec{x} , i.e., $\hat{\vec{x}}$ such that $\vec{x} = \hat{\vec{x}} + \delta \vec{x}$

$$\dot{\vec{x}} = \dot{\vec{x}} + \delta \dot{\vec{x}} = f(\hat{\vec{x}} + \delta \vec{x}, t)$$
 (23)

Using Taylor series expansion

$$f(\hat{\vec{x}} + \delta \vec{x}, t) = \dot{\hat{\vec{x}}} + \delta \dot{\vec{x}} = f(\hat{\vec{x}}, t) + \frac{\partial f(\vec{x}, t)}{\partial \vec{x}} \Big|_{\vec{x} = \hat{\vec{x}}} \delta \vec{x} + H.O.T$$

$$\approx \dot{\hat{\vec{x}}} + \frac{\partial f(\vec{x}, t)}{\partial \vec{x}} \Big|_{\vec{x} = \hat{\vec{x}}} \delta \vec{x}$$

$$\Rightarrow \delta \dot{\vec{x}} \approx \left. \frac{\partial f(\vec{x}, t)}{\partial \vec{x}} \right|_{\vec{x} = \hat{\vec{x}}} \delta \vec{x} \tag{24}$$

17 / 23

Actual Measurements



Initially the accelerometer and gyroscope measurements, $\tilde{\vec{f}}_{ib}^{\ b}$ and $\tilde{\vec{\omega}}_{ib}^{\ b}$, respectively, will be modeled as

$$\tilde{\vec{f}}_{ib}^{\ b} = \vec{f}_{ib}^{\ b} + \Delta \vec{f}_{ib}^{\ b} \tag{25}$$

$$\tilde{\vec{\omega}}_{ib}^{\ b} = \vec{\omega}_{ib}^{\ b} + \Delta \vec{\omega}_{ib}^{\ b} \tag{26}$$

where $\vec{f}_{ib}^{\ b}$ and $\vec{\omega}_{ib}^{\ b}$ are the specific force and angular rates, respectively; and $\Delta \vec{f}_{ib}^{\ b}$ and $\Delta \vec{\omega}_{ib}^{\ b}$ represents the errors. In later lectures we will discuss more detailed description of these errors.

Actual Measurements



Initially the accelerometer and gyroscope measurements, $\tilde{\vec{f}}_{ib}^{\ b}$ and $\tilde{\vec{\omega}}_{ib}^{\ b}$, respectively, will be modeled as

$$\tilde{\vec{f}}_{ib}^{b} = \vec{f}_{ib}^{b} + \Delta \vec{f}_{ib}^{b}$$
these terms may
$$\tilde{\vec{\omega}}_{ib}^{b} = \vec{\omega}_{ib}^{b} + \Delta \vec{\omega}_{ib}^{b}$$
be expanded further
$$(25)$$

$$\tilde{\vec{\omega}}_{ib}^{\ b} = \vec{\omega}_{ib}^{\ b} + \left(\Delta \vec{\omega}_{ib}^{\ b}\right)^{\text{be expanded further}} \tag{26}$$

where $\vec{f}_{ib}^{\ b}$ and $\vec{\omega}_{ib}^{\ b}$ are the specific force and angular rates, respectively; and $\Delta \vec{f}_{ib}^{\ b}$ and $\Delta \vec{\omega}_{ib}^{b}$ represents the errors. In later lectures we will discuss more detailed description of these errors.



Accelerometers

$$\tilde{\vec{f}}_{ib}^{\ b} = \vec{b}_a + (\mathcal{I} + M_a)\vec{f}_{ib}^{\ b} + \vec{nl}_a + \vec{w}_a \tag{27}$$

Gyroscopes

$$\tilde{\vec{\omega}}_{ib}^{\ b} = \vec{b}_g + (\mathcal{I} + M_g)\vec{\omega}_{ib}^{\ b} + G_g\vec{f}_{ib}^{\ b} + \vec{w}_g$$
 (28)



Accelerometers

$$\tilde{\vec{f}}_{ib}^{b} = \vec{b}_{a} + (\mathcal{I} + M_{a})\vec{f}_{ib}^{b} + \vec{nl}_{a} + \vec{w}_{a}$$
Biases
$$\tilde{\vec{\omega}}_{ib}^{b} = \vec{b}_{g} + (\mathcal{I} + M_{g})\vec{\omega}_{ib}^{b} + G_{g}\vec{f}_{ib}^{b} + \vec{w}_{g}$$

Gyroscopes

Inertial Measurements 000 EE 565: Position, Navigation and Timing

(28)



Accelerometers

$$ilde{ec{f}}_{ib}^{b} = ec{b}_{a} + (\mathcal{I} + ec{M_{a}}) ec{f}_{ib}^{b} + ec{n} ec{l}_{a} + ec{w}_{a}$$

Misalignment and SF Errors

Gyroscopes

$$ilde{ec{\omega}}_{ib}^{\ b} = ec{b}_{m{g}} + (\mathcal{I} + ec{\mathcal{M}}_{m{g}}) ec{ec{\omega}}_{ib}^{\ b} + G_{m{g}} ec{f}_{ib}^{\ b} + ec{w}_{m{g}}$$

,

(28)



Accelerometers

$$\vec{ ilde{f}}_{ib}^{\ b} = \vec{b}_a + (\mathcal{I} + M_a) \vec{f}_{ib}^{\ b} + \vec{nl}_a + \vec{w}_a$$

Non-linearity

Gyroscopes

$$\tilde{\vec{\omega}}_{ib}^{b} = \vec{b}_{g} + (\mathcal{I} + M_{g})\vec{\omega}_{ib}^{b} + G_{g}\vec{f}_{ib}^{b} + \vec{w}_{g}$$
 (28)



Accelerometers

$$\tilde{\vec{f}}_{ib}^{\ b} = \vec{b}_a + (\mathcal{I} + M_a)\vec{f}_{ib}^{\ b} + \vec{nl}_a + \vec{w}_a \tag{27}$$

Gyroscopes

$$\tilde{\vec{\omega}}_{ib}^{b} = \vec{b}_g + (\mathcal{I} + M_g)\vec{\omega}_{ib}^{b} + \vec{G}_g\vec{f}_{ib}^{b} + \vec{w}_g$$

G-Sensitivity

(28)



Accelerometers

$$\tilde{\vec{f}}_{ib}^{\ b} = \vec{b}_a + (\mathcal{I} + M_a)\vec{f}_{ib}^{\ b} + \vec{nl}_a + \vec{v}_a$$
Noise

Gyroscopes

$$\tilde{\vec{\omega}}_{ib}^{\ b} = \vec{b}_g + (\mathcal{I} + M_g)\vec{\omega}_{ib}^{\ b} + G_g\vec{f}_{ib}^{\ b} + \vec{v}_g$$
 (28)

Inertial Measurements 000

Pos, Vel, Force and Angular Rate Errors



(32)

20 / 23

Position error

$$\delta \vec{r}_{\beta b}^{\gamma} = \vec{r}_{\beta b}^{\gamma} - \hat{\vec{r}}_{\beta b}^{\gamma} \tag{29}$$

Velocity error

$$\delta \vec{v}_{\beta b}^{\gamma} = \vec{v}_{\beta b}^{\gamma} - \hat{\vec{v}}_{\beta b}^{\gamma} \tag{30}$$

Specific force errors

$$\delta \vec{f}_{ib}^{\ b} = \vec{f}_{ib}^{\ b} - \hat{\vec{f}}_{ib}^{\ b} \tag{31}$$

$$\Delta_e \vec{f}_{ib}^{\ b} = \Delta \vec{f}_{ib}^{\ b} - \Delta \hat{\vec{f}}_{ib}^{\ b} = -\delta \vec{f}_{ib}^{\ b}$$

Angular rate errors

$$\delta \vec{\omega}_{ib}^{\ b} = \vec{\omega}_{ib}^{\ b} - \hat{\vec{\omega}}_{ib}^{\ b} \tag{33}$$

$$\Delta_e \vec{\omega}_{ib}^{\ b} = \Delta \vec{\omega}_{ib}^{\ b} - \Delta \hat{\vec{\omega}}_{ib}^{\ b} = -\delta \vec{\omega}_{ib}^{\ b} \tag{34}$$

Attitude Error Definition



Define

$$\delta C_b^{\gamma} = C_b^{\gamma} \hat{C}_{\gamma}^b = e^{[\delta \vec{\psi}_{\gamma b}^{\gamma} \times]} \approx \mathcal{I} + [\delta \vec{\psi}_{\gamma b}^{\gamma} \times]$$
(35)

This is the error in attitude resulting from errors in estimating the angular rates.

Attitude Error Properties



The attitude error is a multiplicative small angle transformation from the actual frame to the computed frame

$$\hat{C}_{b}^{\gamma} = (\mathcal{I} - [\delta \vec{\psi}_{\gamma b}^{\gamma} \times]) C_{b}^{\gamma}$$
(36)

Similarly,

$$C_b^{\gamma} = (\mathcal{I} + [\delta \vec{\psi}_{\gamma b}^{\gamma} \times]) \hat{C}_b^{\gamma} \tag{37}$$

Specific Force and Agnular Rates



Similarly we can attempt to estimate the specific force and angular rate by applying correction based on our estimate of the error.

$$\hat{\vec{f}}_{ib}^{\ b} = \tilde{\vec{f}}_{ib}^{\ b} - \Delta \hat{\vec{f}}_{ib}^{\ b} \tag{38}$$

$$\hat{\vec{\omega}}_{ib}^{\ b} = \tilde{\vec{\omega}}_{ib}^{\ b} - \Delta \hat{\vec{\omega}}_{ib}^{\ b} \tag{39}$$

where $\hat{\vec{f}}_{ib}^{\ b}$ and $\hat{\vec{\omega}}_{ib}^{\ b}$ are the accelerometer and gyroscope estimated calibration values, respectively.