EE 570: Location and Navigation

Navigation Mathematics: Kinematics (Quaternions)

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Singularity Problem



Euler angles and angle-axis rotation consist of only three elements and they are not unique, e.g., there are orientations that are represented by different Euler angles and angle-axis rotation.

Quaternion

Quaternions are 4-element representation of the rotation vectors. The additional element makes quaternions unique. Since quaternions are only 4 elements, they have the lowest dimentionality possible for a globally nonsignualar attitude representation.

Quaternions



Given a rotation vector defined as

$$\vec{K} = \theta \vec{k}$$

where \vec{k} and θ are Euler axis and angle, respectively, a quaternion is defined as

$$ar{q} = egin{bmatrix} q_s \ ar{q} \end{bmatrix} = egin{bmatrix} q_s \ q_x \ q_y \ q_z \end{bmatrix} = egin{bmatrix} \cos(rac{ heta}{2}) \ ar{k}\sin(rac{ heta}{2}) \end{bmatrix}$$

Quaternion Properties



Quaternion inverse

$$ar{q}^{-1} = egin{bmatrix} q_s \ -q_x \ -q_y \ -q_z \end{bmatrix}$$

Vector transformation Define

$$reve{q} = egin{bmatrix} 0 \ ec{q} \end{bmatrix}$$

then transforming a vector \vec{v}^p defined in the p-frame may be transformed to the i-frame using

$$reve{v}^i = ar{q} \otimes reve{v}^p \otimes ar{q}^{-1}$$

Quaternion Multiply



Quaternion multiply

$$ar{r} = ar{q} \otimes ar{p} = [ar{q} \otimes] ar{p} = egin{bmatrix} q_{s}p_{s} - ec{q} \cdot ec{p} \ q_{s}ec{p} + p_{s}ec{q} + ec{q} imes ec{p} \end{bmatrix}$$

where

$$[ar{q}\otimes] = egin{bmatrix} q_{s} & -q_{x} & -q_{y} & -q_{z} \ q_{x} & q_{s} & -q_{z} & q_{y} \ q_{y} & q_{z} & q_{s} & -q_{x} \ q_{z} & -q_{y} & q_{x} & q_{s} \end{bmatrix}$$

Quaternion Multiply



Quaternion multiply (corresponds to reverse order DCM)

$$ar{r} = ar{q} \circledast ar{p} = [ar{q} \circledast] ar{p} = egin{bmatrix} q_{s}p_{s} - ar{q} \cdot ar{p} \ q_{s}ar{p} + p_{s}ar{q} - ar{q} imes ar{p} \end{bmatrix}$$

•

$$\bar{q}\otimes\bar{p}=\bar{p}\circledast\bar{q}$$

where

$$egin{aligned} [ar{q} \circledast] = egin{bmatrix} q_s & -q_x & -q_y & -q_z \ q_x & q_s & q_z & -q_y \ q_y & -q_z & q_s & q_x \ q_z & q_y & -q_x & q_s \end{bmatrix} \end{aligned}$$

Quaternion to DCM



$$\mathcal{T}(\bar{q}) = (q_s^2 - |\vec{q}|^2)\mathcal{I} + 2q_s[\vec{q}\times] + 2\vec{q}\vec{q}^T =$$

$$= \begin{bmatrix} q_s^2 + q_x^2 - q_y^2 - q_z^2 & 2(q_xq_y - q_sq_z) & 2(q_xq_z + q_sq_y) \\ 2(q_xq_y + q_sq_z) & q_s^2 - q_x^2 + q_y^2 - q_z^2 & 2(q_yq_z - q_sq_x) \\ 2(q_xq_z - q_sq_y) & 2(q_yq_z + q_sq_x) & q_s^2 - q_x^2 - q_y^2 + q_z^2 \end{bmatrix}$$

The above DCM describes the orientation of the object frame as seen by the reference frame, e.g., (C_b^i)

Quaternion to DCM



$$\mathcal{T}^{T}(\bar{q}) = (q_{s}^{2} - |\vec{q}|^{2})\mathcal{I} - 2q_{s}[\vec{q} \times] + 2\vec{q}\vec{q}^{T} =$$

$$= \begin{bmatrix} q_{s}^{2} + q_{x}^{2} - q_{y}^{2} - q_{z}^{2} & 2(q_{x}q_{y} + q_{s}q_{z}) & 2(q_{x}q_{z} - q_{s}q_{y}) \\ 2(q_{x}q_{y} - q_{s}q_{z}) & q_{s}^{2} - q_{x}^{2} + q_{y}^{2} - q_{z}^{2} & 2(q_{y}q_{z} + q_{s}q_{x}) \\ 2(q_{x}q_{z} + q_{s}q_{y}) & 2(q_{y}q_{z} - q_{s}q_{x}) & q_{s}^{2} - q_{x}^{2} - q_{y}^{2} + q_{z}^{2} \end{bmatrix}$$

The above DCM describes the orientation of the reference frame as seen by the object frame, e.g., (C_i^b)

Quaternion Identities



$$\begin{split} [\bar{q}^{-1}\otimes] &= [\bar{q}\otimes]^{-1} = [\bar{q}\otimes]^T \\ [\bar{q}^{-1}\circledast] &= [\bar{q}\circledast]^{-1} = [\bar{q}\circledast]^T \\ [\bar{q}\otimes] &= e^{\frac{1}{2}[\check{k}\otimes]} = \cos(\theta/2)\mathcal{I} + \frac{1}{2}[\check{k}\otimes]\frac{\sin(\theta/2)}{\theta/2} \\ [\bar{q}\circledast] &= e^{\frac{1}{2}[\check{k}\circledast]} = \cos(\theta/2)\mathcal{I} + \frac{1}{2}[\check{k}\circledast]\frac{\sin(\theta/2)}{\theta/2} \\ [\bar{q}\otimes][\bar{q}\circledast]^{-1} &= [\bar{q}\circledast]^{-1}[\bar{q}\otimes] = \begin{bmatrix} 1 & 0 \\ 0 & \mathcal{T}(\bar{q}) \end{bmatrix} \end{split}$$

Quaternion Identities



$$egin{aligned} ar{q}\otimesar{p}\otimesar{r}&=(ar{q}\otimesar{p})\otimesar{r}&=ar{q}\otimes(ar{p}\otimesar{r})\ ar{q}\circledastar{p}\circledastar{r}&=(ar{q}\circledastar{p})\circledastar{r}&=ar{q}\circledast(ar{p}\circledastar{r})\ ar{(ar{q}\circledastar{p})\otimesar{r}}&
eqar{q}\circledast(ar{p}\otimesar{r})\ ar{q}\circledast(ar{p}\otimesar{r})\ ar{(ar{q}\otimesar{p})}\circledastar{r}&
eqar{q}\otimes(ar{p}\otimesar{r})\ \end{aligned}$$